

**Amendments to the Specification:**

Please replace paragraph [0031] with the following rewritten paragraph:

[0031] The exiting combustion gases enter an upper gas reaction area 510 formed from converging walls 500 of exhaust section B1. In this section, the exiting gases ~~further expand and develop high pressure and temperature, ever continuously expanding and rushing travel~~ toward automatic adjustable gas entry point 520 where the exiting gases then enter a lower gas reaction area 620 formed by diverging walls 600 of exhaust section C1. In lower gas reaction area 620, the exiting gases further increase in pressure and temperature and enter the first stage of a multiple stage compound exhaust system 700 provided at section D1 of the exhaust system. As shown, there are three stages formed by stage sections 710, 720 and 730. Continued flow paths of the exiting gases develop multiple action and reaction forces, acting to further extract kinetic force from the gases and further providing thrust force to propel the jet and associated craft upward. A suitable exemplary multiple stage compound exhaust system is the 3-stage compound exhaust system disclosed in U.S. Patent No. 6,367,739, the subject matter of which is hereby incorporated herein by reference in its entirety. However, advantages can be achieved by as few as two stages and as many as 10 or more, the higher the number the higher the efficiency.

Please replace paragraph [0039] with the following rewritten paragraph:

[0039] During operation, quantum jet turbine engines 1200 are started by using either pure distilled water or superchilled air individually or jointly as a propulsion source. Both shown quantum jet engines 1200 will have their upper combustion chambers 1230A isolated from the lower chambers 1230B by locking of gas valve locking devices 1290 provided between the upper and lower combustion chambers. At this time, batteries 2040 are activated to raise the temperature of heating elements 1280 to between 200-400°C or more preferably, ~~in the range of 1000°-3500°C, most preferably~~ between 1000°-2500°C.

Please replace paragraph [0043] with the following rewritten paragraph:

[0043] After passing turbine 1300, the exhaust gases pass through upper gas expansion area 1510 defined by converging walls 1500 of exhaust section B1. In this section, the exiting gases further expand and develop high pressure and temperature ~~ever continuously expanding and rushing~~due to the presence of heating elements 1280 provided in this region and rush toward automatic adjustable gas entry point 1520 where the exiting gases then enter a lower gas reaction area 1620, formed by diverging walls 1600 of exhaust section C1. In lower gas reaction area 1620, the exiting gases further increase in pressure and temperature and enter the first stage of a multiple stage compound exhaust system 1700 provided at section D1 of the exhaust system. As shown, there are three stages. Continued flow paths of the exiting gases develop multiple action and reaction forces, acting to further extract kinetic force from the gases and further providing thrust force to propel the jet and associated craft upward. As in the previous example, a suitable exemplary multiple stage compound exhaust system is the 3-stage compound exhaust system disclosed in U.S. Patent No. 6,367,739, the subject matter of which is hereby incorporated herein by reference in its entirety.